

A novel four-unknown quasi-3D shear deformation theory for functionally graded plates

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(Received March 30, 2017, Revised January 16, 2018, Accepted March 29, 2018)

Abstract. In this article a four unknown quasi-3D shear deformation theory for the bending analysis of functionally graded (FG) plates is developed. The advantage of this theory is that, in addition to introducing the thickness stretching impact ($\varepsilon_z \neq 0$), the displacement field is modeled with only four variables, which is even less than the first order shear deformation theory (FSDT). The principle of virtual work is utilized to determine the governing equations. The obtained numerical results from the proposed theory are compared with the CPT, FSDT, and other quasi-3D HSDTs.

Keywords: quasi 3D theory; bending; functionally graded plate

1. Introduction

Functionally graded materials (FGMs) are a type of advanced composite materials which was initially proposed in Japan (Bever and Duwez 1972, Koizumi 1993). The important feature of this type of composites is the continuity along a desired direction within a structural element such as shell, plate or beam (Matsunaga 2008, Hosseini-Hashemi *et al.* 2010, Reddy 2011, Eltaher *et al.* 2013, Swaminathan and Naveenkumar 2014, Bousahla *et al.* 2014, Yaghoobi *et al.* 2014, Kar and Panda 2015, Darılmaz 2015, Ait Atmane *et al.* 2015, Akbaş 2015, Al-Basyouni *et al.* 2015, Kolahchi *et al.* 2015, Pradhan and Chakraverty 2015, Meradjah *et al.* 2015, Bourada *et al.* 2015, Akbaş 2016, Celebi *et al.* 2016, Bellifa *et al.* 2016, Ghorbanpour Arani *et al.* 2016, Ahouel *et al.* 2016, Raminnea *et al.* 2016, Bellifa *et al.* 2017a, Benadouda *et al.* 2017, Bouafia *et al.* 2017a, Rahmani *et al.* 2017, Aldousari 2017, Bouakal *et al.* 2017, Bakhadda *et al.* 2018).

Some kinds of conventional composites suffer in continuity within the thickness direction; such discontinuity can be adjusted by a gradual and smooth variation of mechanical properties within the thickness of the structural element as in FGMs. Moreover, FGMs allow us to obtain high thermal and toughness mechanical characteristics, as a result of mixing for example ceramic and metal (Tounsi *et al.* 2013, Boudierba *et al.* 2013 and 2016, Zidi *et al.* 2014, Taibi *et al.* 2015, Hamidi *et al.* 2015, Beldjelili *et al.* 2016, Bousahla *et al.* 2016, El-Haina *et al.* 2017, Khetir *et al.*

2017, Menasria *et al.* 2017, Mouffoki *et al.* 2017).

General observations on FGMs can be consulted in the review article by Jha *et al.* (2013). However, a detailed review that focuses on the stress, dynamic and buckling investigation of FG plates was presented by Swaminathan *et al.* (2015). From this article a very large list of studies on shear deformation theories of plates made of FGMs can be found. Both analytical and numerical formulation of the shear deformation plate theories under several types of loads were indicated but without considering the mathematical implication of the employed methodologies and contributions.

The first non-polynomial HSDT was proposed by Levy (1877) and after more than one century the sinusoidal HSDT were investigated and evaluated by Stein (1986). Currently, this HSDT was extensively employed by Touratier (1991), Vidal and Polit (2008, 2009, 2011, 2013), Ghugal (2010), Baseri *et al.* (2016), etc. Then, Soldatos (1992) developed a hyperbolic shear strain shape function; Karama *et al.* (2003) proposed an exponential expansion, etc. Others non-polynomial shear strain shape functions are presented also in the article by Viola *et al.* (2013a, b). Kolahchi and Bidgoli (2016) employed a sinusoidal beam model for dynamic instability of single-walled carbon nanotubes. Arani and Kolahchi (2016) investigated buckling behavior of embedded concrete columns armed with carbon nanotubes. Kolahchi *et al.* (2016a) used differential cubature and quadrature-Bolotin methods for dynamic stability of embedded piezoelectric nanoplates based on visco-nonlocal-piezoelectricity theories. Bilouei *et al.* (2016) examined the stability of concrete columns retrofitted with nano-fiber reinforced polymer. Madani *et al.* (2016) proposed a differential cubature method for vibration

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analysis of embedded FG-CNT-reinforced piezoelectric cylindrical shells subjected to uniform and non-uniform temperature distributions. Kolahchi *et al.* (2016b) analyzed the dynamic stability of temperature-dependent functionally graded CNT-reinforced visco-plates resting on orthotropic elastomeric medium. Kolahchi *et al.* (2017) presented visco-nonlocal-refined Zigzag models for dynamic buckling of laminated nanoplates using differential cubature-Bolotin methods. Zamanian *et al.* (2017) discussed the agglomeration effects on the buckling behavior of embedded concrete columns reinforced with SiO₂ nanoparticles.

A refined and generalized self-consistent model was presented by Bian *et al.* (2005). This is an extension of Soldatos' HSDT (Soldatos 1992) to examine the cylindrical bending response of FG plates. Mantari *et al.* (2012a, b, c, 2014), and Mantari and Granados (2015a, b) proposed several refined non-polynomial HSDTs to investigate the bending, dynamic and thermoelasticity problems of classical composites and FG plates. Mahi *et al.* (2015) developed a new hyperbolic HSDT with five unknowns without including the thickness stretching effect to investigate the static and dynamic analysis of isotropic, functionally graded, sandwich and laminated composite plates. Nami and Janghorban (2013) analyzed the bending response of rectangular nanoplates using trigonometric shear deformation theory based on nonlocal elasticity theory. Akavci (2014) presented an efficient hyperbolic HSDT for free vibration of FG thick rectangular plates on elastic foundation. Hebbali *et al.* (2014) proposed a quasi-3D hyperbolic HSDT for static and vibration analysis of FG plates. Ahmed (2014) studied the post-buckling behavior of sandwich beams with functionally graded faces using a consistent higher order theory. Belabed *et al.* (2014) developed an efficient and simple HSDT for bending and dynamic analyses of FG plates. The number of variables and governing equations were reduced to five instead of six by splitting the transverse displacement into bending, shear and thickness stretching parts. Akavci (2016) proposed a new quasi-3D hyperbolic HSDT for analyzing bending stresses, natural frequencies and buckling loads of FG sandwich plates. Mantari (2016) used a generalized non-polynomial quasi-3D shear deformation theory for advanced composite plates. Liu *et al.* (2017) presented an analysis of FG plates by a simple locking-free quasi-3D hyperbolic plate isogeometric method. Bennoun *et al.* (2016) proposed a novel five variable refined plate theory for vibration analysis of functionally graded sandwich plates. Abdelaziz *et al.* (2017) developed an efficient hyperbolic shear deformation theory for bending, buckling and dynamic of FG sandwich plates with various boundary conditions. Hachemi *et al.* exposed a new simple three-unknown shear deformation theory for bending analysis of FG plates resting on elastic foundations (2017). Sekkal *et al.* (2017b) presented a new quasi-3D HSDT for buckling and vibration of FG plate. Abualnour *et al.* (2018) proposed a novel quasi-3D trigonometric plate theory for free vibration analysis of advanced composite plates. Other works can be also consulted in the articles by Attia *et al.* (2015 and 2018), Belabed *et al.* (2018), Benchohra *et al.*

(2018), Bouhadra *et al.* (2018) and Meksi *et al.* (2018).

In this work, a novel generation of 4-unknown quasi-3D shear deformation theory is proposed. The advantage of this theory is that, in addition to incorporating the thickness stretching influence ($\varepsilon_z \neq 0$), the displacement field is modeled with only 4 unknowns, which is even less than the FSDT and do not need shear correction factor. Thus, the novelty of this work is the use of 4-unknown quasi-3D shear deformation for bending analysis of FG plates, resulting in considerably lower computational effort when compared with the other higher-order theories reported in the literature having more number of unknown functions. The principle of virtual work is utilized to obtain the governing equations. Analytical results from the novel theory are compared with the CPT, FSDT, and other quasi-3D HSDTs. This theory is as accurate as other quasi-3D HSDTs with higher number of variables and so deserves attention.

2. Theoretical formulation

A rectangular plate of uniform thickness h , length a , and width b , made of a FGM is presented in Fig. 1.

The rectangular Cartesian coordinate system x, y, z , has the plane xy at $z = 0$, coinciding with the mid-surface of the plate. The material characteristics can change through the thickness according to the function $V(z)$ as given in the following equation

$$P(z) = \begin{cases} P_b V(z) & V(z) = e^{p\left(\frac{z}{h} + \frac{1}{2}\right)}, \quad \text{Case 1 (exponentially graded)} \\ (P_t - P_b)V(z) + P_b & V(z) = \left(\frac{z}{h} + \frac{1}{2}\right)^p, \quad \text{Case 2 (powerly graded)} \end{cases} \quad (1)$$

where P_t and P_b present the property of the top and bottom faces of the plate, respectively, and p is the exponent that specifies the material distribution profile within the thickness. In this work, for example, the Young's modulus, E , and shear modulus, G , change depending on the case

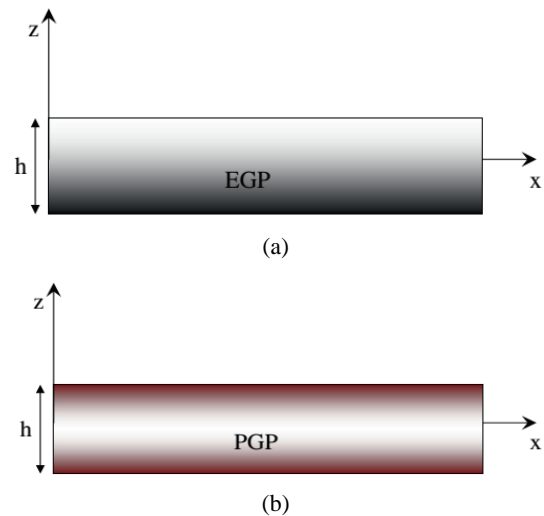


Fig. 1 Geometry of a functionally graded plate

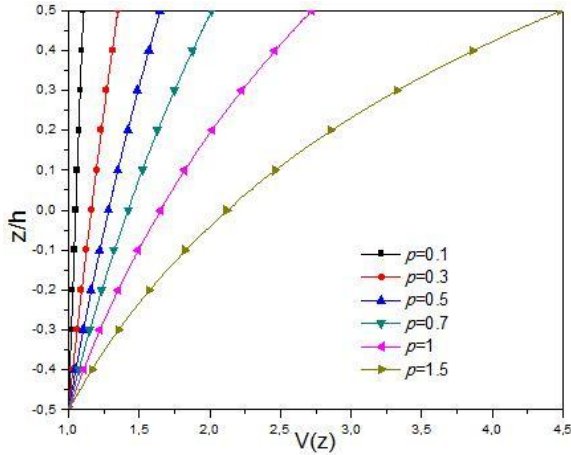


Fig. 2 Exponentially graded function $V(z)$ along the thickness of an EGP for different values of the material parameter p

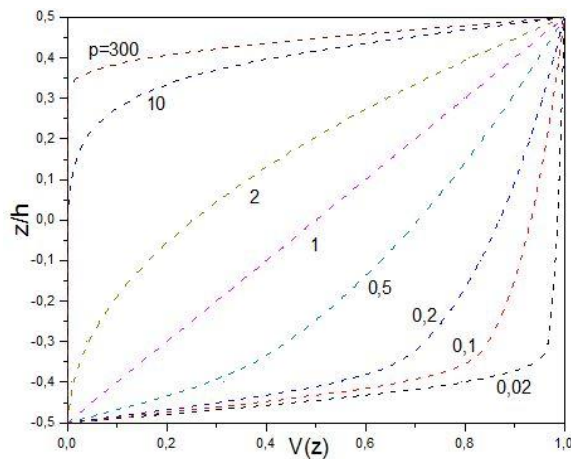


Fig. 3 Powerly graded function $V(z)$ along the thickness of an PGP for different values of the material parameter p

problem according to Eq. (1), and the Poisson ratio, ν considered to be constant. Fig. 2 indicates the exponential function $V(\bar{z} = z/h)$ along the thickness of an exponentially graded plate (EGP) for different values of the material parameter p . While Fig. 3, indicates the corresponding function for powerly graded plates (PGP).

The displacement field satisfying the conditions of transverse shear stresses (and hence strains) vanishing at a point $(x, y, \pm h/2)$ on the top and bottom surfaces of the plate, is expressed as follows (Besseghier *et al.* 2017, Fahsi *et al.* 2017, Zine *et al.* 2018)

$$u(x, y, z) = u_0(x, y) - z \frac{\partial w_0}{\partial x} + k_1 f(z) \int \theta(x, y) dx \quad (2a)$$

$$v(x, y, z) = v_0(x, y) - z \frac{\partial w_0}{\partial y} + k_2 f(z) \int \theta(x, y) dy \quad (2b)$$

$$w(x, y, z) = w_0(x, y) + g(z)\theta(x, y) \quad (2c)$$

The coefficients k_1 and k_2 depends on the geometry. It

can be seen that the kinematic in Eq. (2) introduces only four unknowns (u_0, v_0, w_0 and θ).

In this work, the shape function is taken based on the hyperbolic function given by Mantari and Guedes Soares (2014) as

$$f(z) = \frac{4h}{5} \sinh\left(\frac{5}{4h}z\right) + z \left[-\cosh\left(\frac{5}{8}\right) + \frac{3}{20} \cos\left(\frac{5}{8}\right) \right] \quad (3)$$

$$g(z) = -\frac{3}{20} \cos\left(\frac{5}{4h}z\right)$$

The strain-displacement expressions, based on this formulation, are given as follows

$$\begin{Bmatrix} \varepsilon_x \\ \varepsilon_y \\ \gamma_{xy} \end{Bmatrix} = \begin{Bmatrix} \varepsilon_x^0 \\ \varepsilon_y^0 \\ \gamma_{xy}^0 \end{Bmatrix} + z \begin{Bmatrix} k_x^b \\ k_y^b \\ k_{xy}^b \end{Bmatrix} + f(z) \begin{Bmatrix} k_x^s \\ k_y^s \\ k_{xy}^s \end{Bmatrix} \quad (4)$$

$$\begin{Bmatrix} \gamma_{yz} \\ \gamma_{xz} \end{Bmatrix} = f'(z) \begin{Bmatrix} \gamma_{yz}^0 \\ \gamma_{xz}^0 \end{Bmatrix} + g(z) \begin{Bmatrix} \gamma_{yz}^1 \\ \gamma_{xz}^1 \end{Bmatrix}, \quad \varepsilon_z = g'(z) \varepsilon_z^0$$

Where

$$\begin{Bmatrix} \varepsilon_x^0 \\ \varepsilon_y^0 \\ \gamma_{xy}^0 \end{Bmatrix} = \begin{Bmatrix} \frac{\partial u_0}{\partial x} \\ \frac{\partial v_0}{\partial x} \\ \frac{\partial u_0}{\partial y} + \frac{\partial v_0}{\partial x} \end{Bmatrix}, \quad \begin{Bmatrix} k_x^b \\ k_y^b \\ k_{xy}^b \end{Bmatrix} = \begin{Bmatrix} -\frac{\partial^2 w_0}{\partial x^2} \\ -\frac{\partial^2 w_0}{\partial y^2} \\ -2 \frac{\partial^2 w_0}{\partial x \partial y} \end{Bmatrix}, \quad (5a)$$

$$\begin{Bmatrix} k_x^s \\ k_y^s \\ k_{xy}^s \end{Bmatrix} = \begin{Bmatrix} k_1 \theta \\ k_2 \theta \\ k_1 \frac{\partial}{\partial y} \int \theta dx + k_2 \frac{\partial}{\partial x} \int \theta dy \end{Bmatrix}$$

$$\begin{Bmatrix} \gamma_{yz}^0 \\ \gamma_{xz}^0 \end{Bmatrix} = \begin{Bmatrix} k_2 \int \theta dy \\ k_1 \int \theta dx \end{Bmatrix}, \quad \begin{Bmatrix} \gamma_{yz}^1 \\ \gamma_{xz}^1 \end{Bmatrix} = \begin{Bmatrix} \frac{\partial \theta}{\partial y} \\ \frac{\partial \theta}{\partial x} \end{Bmatrix}, \quad \varepsilon_z^0 = \theta \quad (5b)$$

and

$$g'(z) = \frac{dg(z)}{dz} \quad (5c)$$

The integrals presented in the above equations shall be resolved by a Navier type method and can be expressed as follows

$$\frac{\partial}{\partial y} \int \theta dx = A' \frac{\partial^2 \theta}{\partial x \partial y}, \quad \frac{\partial}{\partial x} \int \theta dy = B' \frac{\partial^2 \theta}{\partial x \partial y}, \quad (6)$$

$$\int \theta dx = A' \frac{\partial \theta}{\partial x}, \quad \int \theta dy = B' \frac{\partial \theta}{\partial y}$$

where the coefficients A' and B' are considered according to the type of solution employed, in this case via Navier method. Therefore, $A'B', k_1$ and k_2 are expressed as follows

$$A' = -\frac{1}{\alpha^2}, \quad B' = -\frac{1}{\beta^2}, \quad k_1 = -\alpha^2, \quad k_2 = -\beta^2 \quad (7)$$

where α and β are defined in expression (20).

The linear constitutive relations are given below

$$\begin{Bmatrix} \sigma_x \\ \sigma_y \\ \sigma_z \\ \tau_{xy} \\ \tau_{xz} \\ \tau_{yz} \end{Bmatrix} = \begin{bmatrix} C_{11} & C_{12} & C_{13} & 0 & 0 & 0 \\ C_{12} & C_{22} & C_{23} & 0 & 0 & 0 \\ C_{13} & C_{23} & C_{33} & 0 & 0 & 0 \\ 0 & 0 & 0 & C_{66} & 0 & 0 \\ 0 & 0 & 0 & 0 & C_{55} & 0 \\ 0 & 0 & 0 & 0 & 0 & C_{44} \end{bmatrix} \begin{Bmatrix} \varepsilon_x \\ \varepsilon_y \\ \varepsilon_z \\ \gamma_{xy} \\ \gamma_{xz} \\ \gamma_{yz} \end{Bmatrix} \quad (8)$$

where C_{ij} are the three-dimensional elastic constants defined by

$$C_{11} = C_{22} = C_{33} = \frac{(1-\nu)E(z)}{(1-2\nu)(1+\nu)}, \quad (9a)$$

$$C_{12} = C_{13} = C_{23} = \frac{\nu E(z)}{(1-2\nu)(1+\nu)}, \quad (9b)$$

$$C_{44} = C_{55} = C_{66} = \frac{E(z)}{2(1+\nu)}, \quad (9c)$$

Considering the static version of the principle of virtual work, the following relations can be obtained (Fekrar *et al.* 2014, Ait Amar Meziane *et al.* 2014, Larbi Chaht *et al.* 2015, Ait Yahia *et al.* 2015, Belkorissat *et al.* 2015, Zemri *et al.* 2015, Boukhari *et al.* 2016, Houari *et al.* 2016, Bounouara *et al.* 2016, Draiche *et al.* 2016, Zidi *et al.* 2017, Bellifa *et al.* 2017b, Chikh *et al.* 2017, Klouche *et al.* 2017, Kaci *et al.* 2018, Yazid *et al.* 2018, Youcef *et al.* 2018, Mokhtar *et al.* 2018)

$$0 = \left[\int_{-h/2}^{h/2} \left\{ \int_{\Omega} \left[\sigma_x \delta \varepsilon_x + \sigma_y \delta \varepsilon_y + \sigma_z \delta \varepsilon_z + \tau_{xy} \delta \gamma_{xy} + \tau_{yz} \delta \gamma_{yz} + \tau_{xz} \delta \gamma_{xz} \right] dxdy \right\} dz \right] - \left[\int_{\Omega} q \delta w dxdy \right] \quad (10)$$

$$0 = \int_{\Omega} \left[\begin{aligned} & N_x \delta \varepsilon_x^0 + N_y \delta \varepsilon_y^0 + N_z \delta \varepsilon_z^0 + N_{xy} \delta \gamma_{xy}^0 \\ & + M_x^b \delta k_x^b + M_y^b \delta k_y^b + M_{xy}^b \delta k_{xy}^b + M_x^s \delta k_x^s \\ & + M_y^s \delta k_y^s + M_{xy}^s \delta k_{xy}^s + Q_{yz}^s \delta \gamma_{yz}^0 + S_{yz}^s \delta \gamma_{yz}^1 \\ & + Q_{xz}^s \delta \gamma_{xz}^0 + S_{xz}^s \delta \gamma_{xz}^1 - q \delta w \end{aligned} \right] dxdy \quad (11)$$

Where Ω is the top surface and the stress resultants N , M , S and Q are defined by

$$\begin{aligned} (N_i, M_i^b, M_i^s) &= \int_{-h/2}^{h/2} (1, z, f) \sigma_i dz, \\ (N_i, M_i^b, M_i^s) &= \int_{-h/2}^{h/2} (1, z, f) \sigma_i dz, \quad (i = x, y, xy); \\ N_z &= \int_{-h/2}^{h/2} g'(z) \sigma_z dz \end{aligned} \quad (12a)$$

and

$$\begin{aligned} (S_{xz}^s, S_{yz}^s) &= \int_{-h/2}^{h/2} g(\tau_{xz}, \tau_{yz}) dz, \\ (Q_{xz}^s, Q_{yz}^s) &= \int_{-h/2}^{h/2} f'(\tau_{xz}, \tau_{yz}) dz \end{aligned} \quad (12b)$$

The static version of the governing equations are obtained from Eq. (11) by integrating the displacement gradients by parts and setting the coefficients of δu_0 , δv_0 , δw_0 and $\delta \theta$ to zero separately. The generalized equations obtained are as follows

$$\begin{aligned} \delta u_0: \quad & \frac{\partial N_x}{\partial x} + \frac{\partial N_{xy}}{\partial y} = 0 \\ \delta v_0: \quad & \frac{\partial N_{xy}}{\partial x} + \frac{\partial N_y}{\partial y} = 0 \\ \delta w_0: \quad & \frac{\partial^2 M_x^b}{\partial x^2} + 2 \frac{\partial^2 M_{xy}^b}{\partial x \partial y} + \frac{\partial^2 M_y^b}{\partial y^2} = -q \\ \delta \theta: \quad & -k_1 A' \frac{\partial^2 M_x^s}{\partial x^2} - k_2 B' \frac{\partial^2 M_y^s}{\partial y^2} - (k_1 A' + k_2 B') \frac{\partial^2 M_{xy}^s}{\partial x \partial y} - N_z \\ & + k_1 A' \frac{\partial Q_{xz}^s}{\partial x} + k_2 B' \frac{\partial Q_{yz}^s}{\partial y} + \frac{\partial S_{xz}^s}{\partial x} + \frac{\partial S_{yz}^s}{\partial y} = \frac{3}{20} \cos\left(\frac{5}{8}\right) q \end{aligned} \quad (13)$$

Substituting Eq. (5) into Eq. (8) and the subsequent results into Eqs. (12), the stress resultants are obtained in terms of strains as following compact form

$$\begin{aligned} \begin{Bmatrix} N \\ M^b \\ M^s \end{Bmatrix} &= \begin{bmatrix} A & B & B^s \\ B & D & D^s \\ B^s & D^s & H^s \end{bmatrix} \begin{Bmatrix} \varepsilon \\ k^b \\ k^s \end{Bmatrix} + \theta \begin{Bmatrix} L \\ L^a \\ R \end{Bmatrix}, \\ \begin{Bmatrix} Q \\ S \end{Bmatrix} &= \begin{bmatrix} F^s & X^s \\ X^s & A^s \end{bmatrix} \begin{Bmatrix} \gamma^0 \\ \gamma^1 \end{Bmatrix} \end{aligned} \quad (14a)$$

$$N_z = L(\varepsilon_x^0 + \varepsilon_y^0) + L^a(k_x^b + k_y^b) + R(k_x^s + k_y^s) + R^a \varepsilon_z^0 \quad (14b)$$

in which

$$\begin{aligned} N &= \{N_x, N_y, N_{xy}\}^t, \quad M^b = \{M_x^b, M_y^b, M_{xy}^b\}^t, \\ M^s &= \{M_x^s, M_y^s, M_{xy}^s\}^t \end{aligned} \quad (15a)$$

$$\begin{aligned} \varepsilon &= \{\varepsilon_x^0, \varepsilon_y^0, \gamma_{xy}^0\}^t, \quad k^b = \{k_x^b, k_y^b, k_{xy}^b\}^t, \\ k^s &= \{k_x^s, k_y^s, k_{xy}^s\}^t \end{aligned} \quad (15b)$$

$$\begin{aligned} A &= \begin{bmatrix} A_{11} & A_{12} & 0 \\ A_{12} & A_{22} & 0 \\ 0 & 0 & A_{66} \end{bmatrix}, \quad B = \begin{bmatrix} B_{11} & B_{12} & 0 \\ B_{12} & B_{22} & 0 \\ 0 & 0 & B_{66} \end{bmatrix}, \\ D &= \begin{bmatrix} D_{11} & D_{12} & 0 \\ D_{12} & D_{22} & 0 \\ 0 & 0 & D_{66} \end{bmatrix} \end{aligned} \quad (15c)$$

$$B^s = \begin{bmatrix} B_{11}^s & B_{12}^s & 0 \\ B_{12}^s & B_{22}^s & 0 \\ 0 & 0 & B_{66}^s \end{bmatrix}, \quad D^s = \begin{bmatrix} D_{11}^s & D_{12}^s & 0 \\ D_{12}^s & D_{22}^s & 0 \\ 0 & 0 & D_{66}^s \end{bmatrix}, \quad (15d)$$

$$H^s = \begin{bmatrix} H_{11}^s & H_{12}^s & 0 \\ H_{12}^s & H_{22}^s & 0 \\ 0 & 0 & H_{66}^s \end{bmatrix}$$

$$Q = \{Q_{xz}^s, Q_{yz}^s\}^t, \quad S = \{S_{xz}^s, S_{yz}^s\}^t, \quad (15e)$$

$$\gamma^0 = \{\gamma_{xz}^0, \gamma_{yz}^0\}^t, \quad \gamma^1 = \{\gamma_{xz}^1, \gamma_{yz}^1\}^t$$

$$F^s = \begin{bmatrix} F_{55}^s & 0 \\ 0 & F_{44}^s \end{bmatrix}, \quad X^s = \begin{bmatrix} X_{55}^s & 0 \\ 0 & X_{44}^s \end{bmatrix}, \quad A^s = \begin{bmatrix} A_{55}^s & 0 \\ 0 & A_{44}^s \end{bmatrix} \quad (15f)$$

$$\begin{Bmatrix} L \\ L^a \\ R \\ R^a \end{Bmatrix} = \int_z \lambda(z) \begin{Bmatrix} 1 \\ z \\ f(z) \\ g'(z) \frac{1-\nu}{\nu} \end{Bmatrix} g'(z) dz \quad (15g)$$

and stiffness components are given as

$$\begin{Bmatrix} A_{11} & B_{11} & D_{11} & B_{11}^s & D_{11}^s & H_{11}^s \\ A_{12} & B_{12} & D_{12} & B_{12}^s & D_{12}^s & H_{12}^s \\ A_{66} & B_{66} & D_{66} & B_{66}^s & D_{66}^s & H_{66}^s \end{Bmatrix} = \int_z \lambda(z) \begin{Bmatrix} 1 \\ z \\ z^2 \\ f(z) \\ z f(z) \\ f^2(z) \end{Bmatrix} \begin{Bmatrix} \frac{1-\nu}{\nu} \\ \nu \\ 1-2\nu \\ 2\nu \end{Bmatrix} dz \quad (16a)$$

$$(A_{22}, B_{22}, D_{22}, B_{22}^s, D_{22}^s, H_{22}^s) = (A_{11}, B_{11}, D_{11}, B_{11}^s, D_{11}^s, H_{11}^s) \quad (16b)$$

$$\begin{aligned} & (F_{44}^s, X_{44}^s, A_{44}^s) \\ &= \int_{-h/2}^{h/2} \frac{E(z)}{2(1+\nu)} ([f'(z)]^2, f'(z)g(z), g^2(z)) dz \end{aligned} \quad (16c)$$

$$(F_{55}^s, X_{55}^s, A_{55}^s) = (F_{44}^s, X_{44}^s, A_{44}^s) \quad (16d)$$

Introducing Eq. (14) into Eq. (13), the equations of motion can be expressed in terms of displacements (u_0 , v_0 , w_0 , θ) and the appropriate equations take the form

$$\begin{aligned} & A_{11}d_{11}u_0 + A_{66}d_{22}u_0 + (A_{12} + A_{66})d_{12}v_0 - B_{11}d_{111}w_0 \\ & - (B_{12} + 2B_{66})d_{122}w_0 + (B_{66}^s(k_1A' + k_2B') + B_{12}^s k_2B')d_{122}\theta \\ & + B_{11}^s k_1A'd_{111}\theta + Ld_1\theta = 0, \end{aligned} \quad (17a)$$

$$\begin{aligned} & A_{22}d_{22}v_0 + A_{66}d_{11}v_0 + (A_{12} + A_{66})d_{12}u_0 - B_{22}d_{222}w_0 \\ & - (B_{12} + 2B_{66})d_{112}w_0 + (B_{66}^s(k_1A' + k_2B') + B_{12}^s k_1A')d_{112}\theta \\ & + B_{22}^s k_2B'd_{222}\theta + Ld_2\theta = 0, \end{aligned} \quad (17b)$$

$$\begin{aligned} & B_{11}d_{111}u_0 + (B_{12} + 2B_{66})d_{122}u_0 + (B_{12} + 2B_{66})d_{112}v_0 \\ & + B_{22}d_{222}v_0 - D_{11}d_{1111}w_0 - 2(D_{12} + 2D_{66})d_{1122}w_0 \\ & - D_{22}d_{2222}w_0 + D_{11}^s k_1A'd_{1111}\theta + \\ & ((D_{12}^s + 2D_{66}^s)(k_1A' + k_2B'))d_{1122}\theta + D_{22}^s k_2B'd_{2222}\theta \\ & + L^a(d_{11}\theta + d_{22}\theta) + q = 0 \end{aligned} \quad (17c)$$

$$\begin{aligned} & -k_1A'B_{11}^s d_{111}u_0 - (B_{12}^s k_2B' + B_{66}^s(k_1A' + k_2B'))d_{122}u_0 \\ & - (B_{22}^s k_1A' + B_{66}^s(k_1A' + k_2B'))d_{112}v_0 - B_{22}^s k_2B'd_{222}v_0 \\ & + D_{11}^s k_1A'd_{1111}w_0 + ((D_{12}^s + 2D_{66}^s)(k_1A' + k_2B'))d_{1122}w_0 \\ & + D_{22}^s k_2B'd_{2222}w_0 - H_{11}^s(k_1A')^2 d_{1111}\theta - H_{22}^s(k_2B')^2 d_{2222}\theta \\ & - (2H_{12}^s k_1 k_2 A'B' + (k_1A' + k_2B')^2 H_{66}^s)d_{1122}\theta \\ & + ((k_1A')^2 F_{55}^s + 2k_1A'X_{55}^s + A_{55}^s)d_{11}\theta \\ & + ((k_2B')^2 F_{44}^s + 2k_2B'X_{44}^s + A_{44}^s)d_{22}\theta \\ & - 2R(k_1A'd_{11}\theta + k_2B'd_{22}\theta) - L(d_{11}u_0 + d_{22}v_0) \\ & + L^a(d_{11}w_0 + d_{22}w_0) - R^a\theta - \frac{3}{20}\cos\left(\frac{5}{8}\right)q_m = 0 \end{aligned} \quad (17d)$$

where d_{ij} , d_{ijl} and d_{ijlm} are the following differential operators

$$\begin{aligned} d_i &= \frac{\partial}{\partial x_i}, \quad d_{ij} = \frac{\partial^2}{\partial x_i \partial x_j}, \quad d_{ijl} = \frac{\partial^3}{\partial x_i \partial x_j \partial x_l}, \\ d_{ijlm} &= \frac{\partial^4}{\partial x_i \partial x_j \partial x_l \partial x_m} \quad (i, j, l, m = 1, 2). \end{aligned} \quad (18)$$

The Navier solution method is employed to determine the analytical solutions for which the displacement variables are written as product of arbitrary parameters and known trigonometric functions to respect the equations of motion and boundary conditions.

$$\begin{Bmatrix} u_0 \\ v_0 \\ w_0 \\ \theta \end{Bmatrix} = \sum_{m=1}^{\infty} \sum_{n=1}^{\infty} \begin{Bmatrix} U_{mn} \cos(\alpha x) \sin(\beta y) \\ V_{mn} \sin(\alpha x) \cos(\beta y) \\ W_{mn} \sin(\alpha x) \sin(\beta y) \\ X_{mn} \sin(\alpha x) \sin(\beta y) \end{Bmatrix} \quad (19)$$

with

$$\alpha = m\pi/a \quad \text{and} \quad \beta = n\pi/b \quad (20)$$

The transverse load q is also expanded in the double-Fourier sine series as

$$Q(x, y) = \sum_{m=1}^{\infty} \sum_{n=1}^{\infty} q \sin(\alpha x) \sin(\beta y) \quad (21)$$

Substituting Eq. (19) into Eq. (17), the following problem is obtained

$$\begin{Bmatrix} S_{11} & S_{12} & S_{13} & S_{14} \\ S_{12} & S_{22} & S_{23} & S_{24} \\ S_{13} & S_{23} & S_{33} & S_{34} \\ S_{14} & S_{24} & S_{34} & S_{44} \end{Bmatrix} \begin{Bmatrix} U_{mn} \\ V_{mn} \\ W_{mn} \\ X_{mn} \end{Bmatrix} = \begin{Bmatrix} 0 \\ 0 \\ q \\ -\frac{3}{20}\cos\left(\frac{5}{8}\right)q \end{Bmatrix} \quad (22)$$

Table 1 Non-dimensional center deflection $\bar{w}(a/2, b/2, 0)$ for various EGP's ($a/h = 2$)

b/a	Theory	$p = 0.1$	$p = 0.5$	$p = 1$	$p = 1.5$
6	3-D (Zenkour 2007)	1.638	1.352	1.059	0.826
	Mantari and Guedes Soares (2014)	1.658	1.354	1.045	0.802
	Mantari and Guedes Soares (2013)	1.637	1.336	1.033	0.794
	Mantari and Guedes Soares (2012d)	1.735	1.418	1.100	0.850
	TPT (Zenkour 2007)	1.629	1.331	1.028	0.791
	HPT (Zenkour 2007)	1.548	1.265	0.980	0.756
	Present	1.658	1.354	1.045	0.802
1	3-D (Zenkour 2007)	1.638	1.352	1.059	0.826
	Mantari and Guedes Soares (2014)	1.658	1.354	1.045	0.820
	Mantari and Guedes Soares (2013)	1.637	1.336	1.033	0.794
	Mantari and Guedes Soares (2012d)	1.735	1.418	1.100	0.850
	TPT (Zenkour 2007)	1.629	1.331	1.028	0.791
	HPT (Zenkour 2007)	1.548	1.265	0.980	0.756
	Present	1.658	1.354	1.045	0.802

Where

$$\begin{aligned}
 S_{11} &= A_{11}\alpha^2 + A_{66}\beta^2, & S_{12} &= \alpha\beta (A_{12} + A_{66}), \\
 S_{13} &= -\alpha(B_{11}\alpha^2 + (B_{12} + 2B_{66})\beta^2), \\
 S_{14} &= \alpha \left((k_2 B' B_{12}^s + (k_1 A' + k_2 B') B_{66}^s) \beta^2 \right. \\
 &\quad \left. + k_1 A' B_{11}^s \alpha^2 - L \right), \\
 S_{22} &= A_{66}\alpha^2 + A_{22}\beta^2, \\
 S_{23} &= -\beta(B_{22}\beta^2 + (B_{12} + 2B_{66})\alpha^2), \\
 S_{24} &= \beta \left((k_1 A' B_{12}^s + (k_1 A' + k_2 B') B_{66}^s) \alpha^2 \right. \\
 &\quad \left. + k_2 B' B_{22}^s \beta^2 - L \right), \\
 S_{33} &= D_{11}\alpha^4 + 2(D_{12} + 2D_{66})\alpha^2\beta^2 + D_{22}\beta^4 \\
 S_{34} &= -k_1 A' D_{11}^s \alpha^4 - 2D_{12}^s k_2 B' \alpha^2 \beta^2 \\
 &\quad - 2D_{66}^s (k_1 A' + k_2 B') \alpha^2 \beta^2 \\
 &\quad - k_2 B' D_{22}^s \beta^4 + L^a (\alpha^2 + \beta^2), \\
 S_{44} &= (k_1 A')^2 H_{11}^s \alpha^4 + (2k_1 k_2 A' B' H_{12}^s \\
 &\quad + (k_1 A' + k_2 B')^2 H_{66}^s) \alpha^2 \beta^2 \\
 &\quad + ((k_1 A')^2 F_{55}^s + 2k_1 A' X_{55}^s + A_{55}^s) \alpha^2 \\
 &\quad + (k_2 B')^2 H_{22}^s \beta^4 + R^a - \\
 &\quad 2R(k_1 A' \alpha^2 + k_2 B' \beta^2) \\
 &\quad + ((k_2 B')^2 F_{44}^s + 2k_2 B' X_{44}^s + A_{44}^s) \beta^2
 \end{aligned} \quad (23)$$

3. Numerical results and discussion

In this section, the bending analysis of FG plates is presented. For this end, various FG plates with different geometry and material characteristics are here studied in order to assess the accuracy of the proposed theory.

Fig. 4 presents the non-dimensional variation of maximum deflection within the plate thickness ($b/a = 1, 2, 3, 4, a/h = 4, p = 0.1$). It can be seen that the maximum

deflection is highly influenced by the aspect ratio (b/a).

For example, FG plates with elastic properties varying exponentially in z , as reported by Zenkour (2007); FG plates with elastic properties powerly graded along the thickness direction z , as indicated by Zenkour (2006) and accurately solved by Carrera *et al.* (2008).

3.1 Exponentially graded plates

The bending analysis is carried out by utilizing aluminum (bottom, Al) graded exponentially within the thickness of a rectangular plate (see Fig. 1(a)). The material characteristics used for calculating the numerical results are

$$E_b = 70 \text{ GPa}, \quad \nu_b = 0.3 \quad (24)$$

The following non-dimensional quantities are employed

$$\begin{aligned}
 \bar{w} &= w \left(\frac{a}{2}, \frac{b}{2}, z \right) \frac{10E_b h^3}{q_0 a^4}, \\
 \bar{\sigma}_x &= \sigma_x \left(\frac{a}{2}, \frac{b}{2}, z \right) \frac{h^2}{q_0 a^2}, & \bar{\sigma}_y &= \sigma_y \left(\frac{a}{2}, \frac{b}{2}, z \right) \frac{h^2}{q_0 a^2}, \\
 \bar{\tau}_{xz} &= \tau_{xz} \left(0, \frac{b}{2}, z \right) \frac{h}{q_0 a}, & \bar{z} &= \frac{z}{h}
 \end{aligned} \quad (25)$$

The bending results in this section are compared with: (a) the 3D exact solutions and a HSDT with stretching influence by Zenkour (2007); (b) a developed quasi-3D HSDT with 6 unknowns (Mantari and Guedes Soares, 2013); and a recent quasi-3D HSDT proposed by Mantari and Guedes Soares (2014).

Tables 1-5 show results of non-dimensional maximum deflection, normal stresses and shears stresses. It is found from the examination of these tables that the computed results are in good agreements with the published ones.

Fig. 5 plots the non-dimensional normal stresses ($\bar{\sigma}_x$)

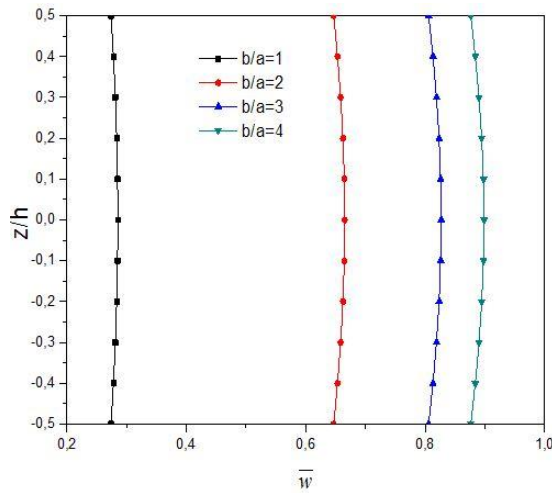


Fig. 4 Variation of non-dimensional displacement, \bar{w} ($a/2$, $b/2$, z), through the thickness of a thick EGP ($a/h = 4$ and $p = 0.5$).

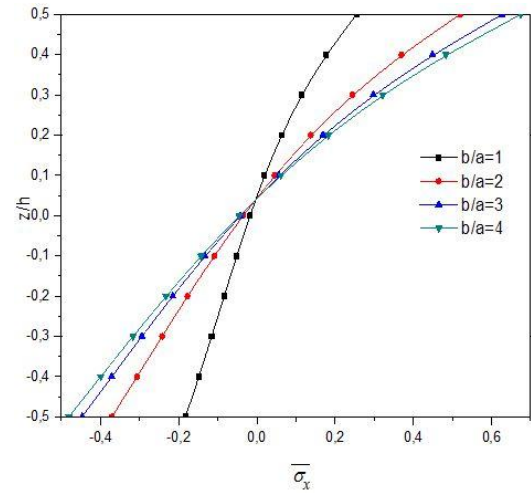


Fig. 5 Variation of non-dimensional normal stress, $\bar{\sigma}_x$ ($a/2$, $b/2$, z), through the thickness of a thick EGP ($a/h = 4$ and $p = 0.5$).

Table 2 Non-dimensional normal stresses $\bar{\sigma}_y$ ($a/2$, $b/2$, $h/2$) for various EGPs ($a/h = 4$)

b/a	Theory	$p = 0.1$	$p = 0.5$	$p = 1$	$p = 1.5$
6	3-D (Zenkour 2007)	0.206	0.231	0.266	0.309
	Mantari and Guedes Soares (2014)	0.218	0.247	0.289	0.337
	Mantari and Guedes Soares (2013)	0.213	0.239	0.280	0.329
	Mantari and Guedes Soares (2012d)	0.201	0.230	0.271	0.319
	TPT (Zenkour 2007)	0.237	0.268	0.314	0.370
	HPT (Zenkour 2007)	0.282	0.322	0.380	0.448
	Present	0.219	0.247	0.286	0.333
1	3-D (Zenkour 2007)	0.217	0.247	0.290	0.340
	Mantari and Guedes Soares (2014)	0.225	0.256	0.302	0.359
	Mantari and Guedes Soares (2013)	0.224	0.255	0.301	0.356
	Mantari and Guedes Soares (2012d)	0.216	0.248	0.293	0.345
	TPT (Zenkour 2007)	0.235	0.268	0.317	0.374
	HPT (Zenkour 2007)	0.241	0.276	0.326	0.385
	Present	0.224	0.255	0.300	0.352

Table 3 Non-dimensional center deflection $\bar{w}(a/2, b/2, 0)$ for various EGPs ($a/h = 10$)

b/a	Theory	$p = 0.1$	$p = 0.5$	$p = 1$	$p = 1.5$	$p = 2$	$p = 2.5$	$p = 3$
6	Ref ^(a)	1.034	0.845	0.655	0.507	0.391	0.302	0.232
	Ref ^(b)	1.035	0.846	0.656	0.507	0.391	0.302	0.232
	Ref ^(c)	1.039	0.852	0.667	0.524	0.412	0.323	0.254
	TPT ^(b)	1.032	0.844	0.654	0.505	0.39	0.301	0.231
	Present	1.034	0.845	0.655	0.507	0.391	0.302	0.232
1	Ref ^(a)	0.279	0.228	0.177	0.137	0.106	0.081	0.063
	Ref ^(b)	0.280	0.229	0.177	0.137	0.106	0.081	0.063
	Ref ^(c)	0.282	0.231	0.181	0.142	0.111	0.087	0.068
	TPT ^(b)	0.279	0.228	0.177	0.137	0.105	0.081	0.062
	Present	0.279	0.228	0.177	0.137	0.106	0.081	0.063

^(a) Mantari and Guedes Soares (2014); ^(b) Mantari and Guedes Soares (2013);

^(c) Mantari and Guedes Soares (2012d)

Table 4 Non-dimensional normal stresses $\bar{\sigma}_y$ ($a/2, b/2, h/2$) for various EGP ($a/h = 10$)

b/a	Theory	$p = 0.1$	$p = 0.5$	$p = 1$	$p = 1.5$	$p = 2$	$p = 2.5$	$p = 3$
6	Ref ^(a)	0.607	0.693	0.817	0.960	1.127	1.322	1.549
	Ref ^(b)	0.601	0.686	0.808	0.951	1.118	1.312	1.539
	Ref ^(c)	0.603	0.688	0.811	0.954	1.120	1.315	1.542
	TPT ^(b)	0.627	0.717	0.845	0.993	1.165	1.364	1.593
	Present	0.607	0.693	0.816	0.960	1.127	1.322	1.549
1	Ref ^(a)	0.210	0.239	0.281	0.329	0.387	0.455	0.534
	Ref ^(b)	0.206	0.234	0.275	0.324	0.382	0.451	0.532
	Ref ^(c)	0.206	0.235	0.277	0.326	0.385	0.450	0.528
	TPT ^(b)	0.220	0.250	0.294	0.346	0.407	0.477	0.560
	Present	0.210	0.238	0.280	0.329	0.386	0.454	0.534

(a) Mantari and Guedes Soares (2014); (b) Mantari and Guedes Soares (2013);

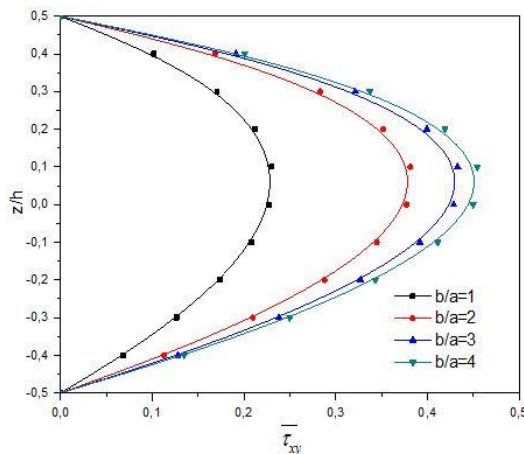
(c) Mantari and Guedes Soares (2012d)

Table 5 Non-dimensional shear stresses $\bar{\tau}_{xz}$ ($0, b/2, 0$) for various EGP ($a/h = 10$)

b/a	Theory	$p = 0.1$	$p = 0.5$	$p = 1$	$p = 1.5$	$p = 2$	$p = 2.5$	$p = 3$
6	Ref ^(a)	0.607	0.693	0.817	0.960	1.127	1.322	1.549
	Ref ^(b)	0.601	0.686	0.808	0.951	1.118	1.312	1.539
	Ref ^(c)	0.603	0.688	0.811	0.954	1.120	1.315	1.542
	TPT ^(b)	0.627	0.717	0.845	0.993	1.165	1.364	1.593
	Present	0.607	0.693	0.816	0.960	1.127	1.322	1.549
1	Ref ^(a)	0.210	0.239	0.281	0.329	0.387	0.455	0.534
	Ref ^(b)	0.206	0.234	0.275	0.324	0.382	0.451	0.532
	Ref ^(c)	0.206	0.235	0.277	0.326	0.385	0.450	0.528
	TPT ^(b)	0.220	0.250	0.294	0.346	0.407	0.477	0.560
	Present	0.210	0.238	0.280	0.329	0.386	0.454	0.534

(a) Mantari and Guedes Soares (2014); (b) Mantari and Guedes Soares (2013);

(c) Mantari and Guedes Soares (2012d)

Fig. 6 Variation of non-dimensional shear stress, $\bar{\tau}_{xz}$ ($0, b/2, z$), through the thickness of a thick EGP ($a/h = 4$ and $p = 0.5$)

within the plate thickness. Again, the computed stress $\bar{\sigma}_x$ is influenced by the aspect ratio (b/a).

Finally, Fig. 6 plots the non-dimensional shear stresses ($\bar{\tau}_{xz}$) variation through the plate thickness.

It can be observed that by increasing the aspect ratio (b/a), the non-dimensional shear stresses are increased.

3.2 Powerly graded plates

A square plate fabricated by metal and ceramic powerly graded within its thickness is presented in Fig. 1(b). In fact, the Young modulus varying in thickness direction of the FG plate according to rule of mixtures is shown in Fig. 3.

$$E_b = 70 \text{ GPa}, \quad E_t = 380 \text{ GPa}, \quad \nu_b = \nu_t = 0.3 \quad (26)$$

The following non-dimensional quantities are employed

$$\begin{aligned} \bar{u} &= u \left(0, \frac{b}{2}, z \right) \frac{100E_t h^3}{qa^4}, & \bar{w} &= w \left(\frac{a}{2}, \frac{b}{2}, z \right) \frac{10E_t h^3}{qa^4}, \\ \bar{\sigma}_x &= \sigma_x \left(\frac{a}{2}, \frac{b}{2}, z \right) \frac{h}{qa}, & \bar{\sigma}_y &= \sigma_y \left(\frac{a}{2}, \frac{b}{2}, z \right) \frac{h}{qa}, \end{aligned} \quad (27)$$

Table 6 Non-dimensional deflection and normal stresses of square PGPs

p	Theory	$\bar{w} (a/2, b/2, 0)$			$\bar{\sigma}_x (a/2, b/2, h/3)$		
		$a/h = 4$	$a/h = 10$	$a/h = 100$	$a/h = 4$	$a/h = 10$	$a/h = 100$
1	Carrera <i>et al.</i> (2008)	0.717	0.588	0.563	0.622	1.506	14.969
	Mantari and Guedes Soares (2014)	0.693	0.568	0.546	0.588	1.459	14.496
	Neves <i>et al.</i> (2011)	0.700	0.585	0.562	0.593	1.495	14.969
	FSDT (Carrera <i>et al.</i> , 2011)	0.729	0.589	0.563	0.806	2.015	20.150
	CPT (Carrera <i>et al.</i> , 2011)	0.562	0.562	0.562	0.806	2.015	20.150
	Present	0.693	0.569	0.546	0.576	1.457	14.482
4	Carrera <i>et al.</i> (2008)	1.159	0.882	0.829	0.488	1.197	11.923
	Mantari and Guedes Soares (2014)	1.092	0.841	0.793	0.434	1.116	11.326
	Neves <i>et al.</i> (2011)	1.118	0.875	0.829	0.440	1.178	11.932
	FSDT (Carrera <i>et al.</i> 2011)	1.113	0.874	0.829	0.642	1.605	16.049
	CPT (Carrera <i>et al.</i> 2011)	0.828	0.828	0.828	0.642	1.605	16.049
	Present	1.092	0.841	0.793	0.417	1.115	11.310
10	Carrera <i>et al.</i> (2008)	1.375	1.007	0.936	0.370	0.897	8.908
	Mantari and Guedes Soares (2014)	1.305	0.979	0.914	0.323	0.836	8.527
	Neves <i>et al.</i> (2011)	1.349	0.875	0.829	0.323	1.178	11.932
	FSDT (Carrera <i>et al.</i> 2011)	1.318	0.997	0.936	0.480	1.199	11.990
	CPT (Carrera <i>et al.</i> 2011)	0.935	0.935	0.935	0.480	1.199	11.990
	Present	1.322	0.978	0.914	0.306	0.836	8.518

Table 7 Non-dimensional displacements and stresses of square PGPs

p	Theory	$\bar{u} (-h/4)$	$\bar{u} (-h/6)$	$\bar{w} (0)$	$\bar{\sigma}_x (h/2)$	$\bar{\sigma}_x (h/3)$	$\bar{\tau}_{yz} (h/6)$	$\bar{\tau}_{xz} (0)$	$\bar{\tau}_{xy} (-h/3)$
1	Ref ^(a)	0.584	0.444	0.568	3.13	1.459	0.299	0.275	0.562
	Ref ^(b)	0.663	0.509	0.589	3.087	1.489	0.262	0.246	0.611
	Present	0.585	0.445	0.569	3.124	1.457	0.291	0.266	0.563
2	Ref ^(a)	0.808	0.629	0.722	3.635	1.344	0.277	0.222	0.494
	Ref ^(b)	0.928	0.731	0.757	3.609	1.395	0.276	0.227	0.544
	Present	0.809	0.630	0.722	3.630	1.342	0.269	0.214	0.495
3	Ref ^(a)	0.907	0.71	0.798	3.876	1.214	0.244	0.185	0.503
	Ref ^(b)	1.045	0.827	0.838	3.874	1.275	0.272	0.211	0.553
	Present	0.908	0.711	0.797	3.870	1.213	0.236	0.178	0.503
5	Ref ^(a)	0.971	0.756	0.872	4.213	1.043	0.195	0.158	0.529
	Ref ^(b)	1.116	0.879	0.912	4.249	1.103	0.243	0.202	0.576
	Present	0.972	0.757	0.871	4.209	1.042	0.189	0.152	0.530
10	Ref ^(a)	1.001	0.762	0.979	5.031	0.836	0.162	0.171	0.554
	Ref ^(b)	1.137	0.876	1.009	5.089	0.878	0.204	0.22	0.589
	Present	1.002	0.762	0.978	5.028	0.836	0.157	0.164	0.554

$$\begin{aligned}
\bar{\tau}_{xy} &= \tau_{xy}(0,0,z) \frac{h}{qa}, & \bar{\tau}_{yz} &= \tau_{yz}\left(\frac{a}{2}, 0, z\right) \frac{h}{qa}, \\
\bar{\tau}_{xz} &= \tau_{xz}\left(0, \frac{b}{2}, z\right) \frac{h}{qa}
\end{aligned} \quad (27)$$

Table 6 shows results of non-dimensional deflection and normal stress results at the specified position (see Eq. (27))

for simply supported homogenous square plate under to bi-sinusoidal distributed load, $p = \{1, 4, 10\}$ and $a/h = 10$. Good results are achieved by the proposed model compared with the ones given by Carrera (2008), Mantari and Guedes Soares (2014), Neves *et al.* (2011) and both FSDT and CPT (Carrera *et al.* 2011).

Table 7 gives results of non-dimensional displacements (\bar{u} and \bar{w}) and normal, in plane shear, and transverse shear

stresses results at the specified position (see Eq. (27)) for simply supported homogenous square plate under to bi-sinusoidal distributed load, $p = \{1, 2, 3, 5, 10\}$ and $a/h = 10$. Good results are achieved by the proposed model compared with the ones given by Mantari and Guedes Soares (2014) and Zenkour (2006). It should be indicated that the theory presented by Zenkour (2006) employs 5 unknowns without considering the thickness stretching influence.

The through thickness variations of displacements and stresses are also illustrated in Fig. 7 for square plates with $a/h = 10$.

4. Conclusions

A novel quasi-3D HSDT with only 4 variables and stretching influences is presented in this work. The governing equations are obtained from the principle of virtual displacements. Analytical solutions are determined for simply supported rectangular plates. By considering further simplifying suppositions to the quasi-3D theory of Zenkour (2007), the number of variables of the novel quasi-3D is diminished by one, and hence, makes the novel theory simple and efficient to utilize. Numerical results demonstrate that these suppositions have a minimal influence on the accuracy of the results for the examined problems. Therefore, it can be deduced that the novel quasi-3D theory is not only accurate but also simple in predicting the bending response of FG plates.

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